6.1.3 Boundary layer and wake quantities — not relevant

6.1.4 Repeatability — see tables B1-2 and B1-3 or B1-10 and B1-11

6.2 Wall interference corrections — no corrections

6.3 Data presentation

6.3.1 Aerodynamic coefficients

6.3.2 Surface pressure coefficients — Cp (x/l) for all sections defined 5.1.1

6.3.3 Flow conditions for

- aerodynamic coefficient data — not relevant

- pressure data — Mo = 0.70/0.84/0.88/0.92 at Re ~ 11.7 10^6 and angles of attack α = 0° to 6°

detail : T2 means table B1-2
F3 means figure B1-3

α \ Mo	0.70	0.84	0.88	0.92
0°	T2/3	T10/11 F3	T18	T25
1°	T4	T12 F4	T19	T26
2°	T5	T13 F5	T20	T27
3°	T6 F10	T14 F6/11	T21 F12	T28 F13
4°	T7	T15 F7	T22	T29
5°	T8	T16 F8	T23	T30
6°	T9	T17 F9	T24	T31

Boundary layer and/or wake data — none

Flow conditions for boundary layer and/or wake data — not relevan

Wall interference corrections included ? — no

Aeroelastic corrections included ? — no

6.3.8 Other corrections ? — no

6.4 Were tests carried out in different facilities on the current model ? If so, what facilities. Are data included in present data base ? — not at transonic Mach numbers

7. References

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8. List of symbols

b	: semi-span
	local chord
c	: mean aerodynamic chord
S	: wing area
	: distance measured along the local chord from the leading-edge of the wing section
y	: distance measured spanwise
	: distance from the plane of the wing
X	: distance measured chordwise from wing apex
	: angle of attack
Mo	: free stream Mach number
M	: local Mach number
Po	: stagnation pressure
po	: free stream static pressure
qo	: free stream dynamic pressure
p	: local static pressure
Cp	: pressure coefficient $Cp = \frac{p - po}{qo}$
To	: stagnation temperature
R_{e_c}	: Reynolds number based on c
CX	: axial force coefficient
CZ	: normal force coefficient
C_l	: rolling moment coefficient
C_m	: pitching moment coefficient
C_n	: yawing moment coefficient